ARMY RESEARCH LABORATORY

Analytical Approximations of Projectile Motion for Linear and Quadratic Air Drag

by Richard Saucier

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The purpose of this report is to provide closed-form solutions to projectile motion while accounting for, or approximating, the effect of air resistance. The plan is to first quickly review the solution when air resistance is neglected, and then to approximate the effect of air drag. Making the assumption that the drag force varies linearly with air speed allows us to solve the equations of motion exactly. The more realistic assumption that air drag varies as the square of the speed can also be solved analytically, provided that we restrict it to shallow launch angles. This analytical approximate solution is compared to the exact numerical solution using a fourth-order Runge-Kutta method.

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1. Introduction

It is well known that projectile motion under the influence of gravity and neglecting air resistance is a problem that can be solved analytically. The closed-form solution is easily manipulated to provide range, maximum range, projectile path, height, time of flight, impact velocity, and angle of impact, among others. In short, we know everything about the motion.

In the more realistic case when air resistance is taken into account, the problem is much harder to deal with (*1, 2*). As a result, the usual strategy is to solve the equations numerically—for example, using the Runge-Kutta method (*3*). This is straightforward enough, but numerical solutions do not provide the same level of insight that we get from analytical formulas. The relatively recent introduction of the Lambert W function (*4*) and its implementation in computer code (*5*) provides a tool that can be applied to this situation (*6, 7*). Although it does not allow us to solve the problem in its full generality, this function does facilitate analytical solutions under reasonable approximations.

The purpose of this report is to provide closed-form solutions to projectile motion while accounting for, or approximating, the effect of air resistance. The plan is to first quickly review the solution when air resistance is neglected, and then approximate the effect of air drag. Making the assumption that the drag force varies linearly with air speed allows us to solve the equations of motion exactly. The more realistic assumption that air drag varies as the square of the speed can also be solved analytically, provided we restrict it to shallow launch angles. This analytical solution is compared to the numerical solution using a fourth-order Runge-Kutta method.

2. Projectile Motion in the Absence of Air Drag

We review the equations of motion for a projectile launched from the ground and falling under the influence of gravity but neglecting air resistance. There is nothing new here, but it does serve to introduce our notation and establish some results that will be useful later.

Thus, let $\mathbf{i}$ and $\mathbf{j}$ be unit vectors along the x (horizontal) and y (vertical) axes, respectively. Then the force acting on the projectile of mass m can be written as $\mathbf{F} = -mg\mathbf{j}$, where g is the acceleration due to gravity. Also, let u be the velocity along the x-axis and w be the velocity along the y-axis, so that $\mathbf{v} = u\mathbf{i} + w\mathbf{j}$ is the total vector velocity of the projectile. Then from Newton's second law, the vector equation of motion,

$$m\frac{d\mathbf{v}}{dt} = -mg\mathbf{j}, \tag{1}$$

can be resolved in terms of its two components,

$$\dot{u} = 0 \quad \text{and} \quad \dot{w} = -g, \tag{2}$$

where dot notation is used to signify derivatives with respect to time. Integrating these equations gives the velocities as a function of time,

$$u = u_0 \quad \text{and} \quad w = w_0 - gt, \tag{3}$$

where u_0 and w_0 are the initial velocities along the x-axis and y-axis, respectively. In terms of the launch angle, θ, as measured from the horizontal, $u_0 = v_0 \cos\theta$ and $w_0 = v_0 \sin\theta$, where v_0 is the launch speed. Integrating again gives the position as a function of time, t,

$$x = (v_0 \cos\theta)\, t \quad \text{and} \quad y = (v_0 \sin\theta)\, t - \tfrac{1}{2} g t^2. \tag{4}$$

Eliminating t between these two equations then gives the equation for the trajectory,

$$y = \left(\frac{\sin\theta}{\cos\theta}\right) x - \left(\frac{g}{2v_0^2 \cos^2\theta}\right) x^2. \tag{5}$$

By completing the square in x, this can also be written as

$$\boxed{y = \frac{v_0^2 \sin^2\theta}{2g} - \frac{g}{2v_0^2 \cos^2\theta}\left(x - \frac{v_0^2 \sin 2\theta}{2g}\right)^2}, \tag{6}$$

which is recognized as the equation for a parabola with its apex (maximum) at $x = (v_0^2 \sin 2\theta)/2g$ and $y = (v_0^2 \sin^2\theta)/2g$. The trajectory is symmetric about this point, which means the range is $(v_0^2 \sin 2\theta)/g$ and the height is $(v_0^2 \sin^2\theta)/2g$. The range, R, can also be obtained directly from equation 5 as the non-trivial value of x that gives $y = 0$:

$$\boxed{R = \frac{v_0^2}{g} \sin 2\theta}. \tag{7}$$

By setting $dR/d\theta = 0$ and solving for θ, or simply by examining $\sin 2\theta$, the maximum range is found to be when the firing angle is 45°, and its value is

$$\boxed{R_{\text{max}} = \frac{v_0^2}{g}}. \tag{8}$$

For a given initial velocity and firing angle, the projectile reaches its maximum height, H, when $w = 0$ and this occurs when $t = v_0 \sin\theta / g$ from equation 3. Substituting this into equation 4 gives

$$\boxed{H = \frac{v_0^2 \sin^2\theta}{2g}}, \tag{9}$$

which agrees with the maximum of the parabolic path found previously. The total time of flight, T, is twice the time to reach its maximum height, so that

$$\boxed{T = \frac{2v_0 \sin\theta}{g}}. \tag{10}$$

The average speed over the entire flight is the range divided by the total time, $\bar{v} = R/T$, which gives

$$\boxed{\bar{v} = v_0 \cos\theta}. \tag{11}$$

Substituting the total flight time into equation 3, the impact velocity is found to be

$$u = v_0 \cos\theta \quad \text{and} \quad w = -v_0 \sin\theta, \tag{12}$$

and it is clear that the following are true:

- the impact angle is the negative of the launch angle, and
- the impact speed is the same as the launch speed.

It is also clear that the motion only depends upon the launch speed, v_0, and launch angle, θ, and is independent of the mass. Typical projectile paths are shown in figure 1. These results serve as points of reference as we consider projectile motion under more realistic conditions of air resistance.

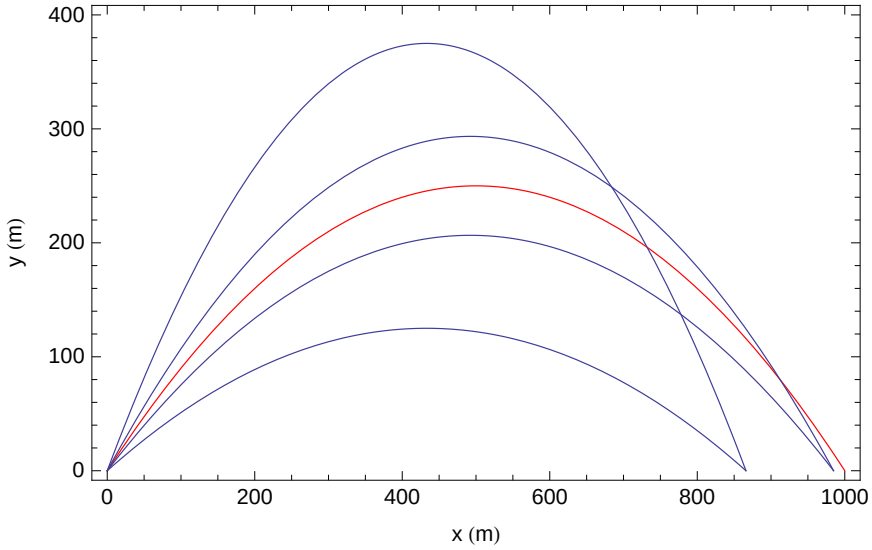

Figure 1. Trajectories neglecting air drag for launch angles of $30°$, $40°$, $50°$, $60°$ (blue) and $\theta_{\max} = 45°$ (red).

3. Projectile Motion with Linear Air Drag ($C_d \sim 1/Ma$)

Now we begin to account for air resistance and determine its effect upon projectile motion. The drag force acting on a projectile due to air resistance is velocity dependent and is of the form

$$\mathbf{F}_d = -\tfrac{1}{2} C_d A \rho v^2 \hat{\mathbf{v}}, \tag{13}$$

where C_d is the (dimensionless) drag coefficient, A is the projectile presented area, ρ is the air density, v is the speed with respect to air, and $\hat{\mathbf{v}}$ is a unit vector in the direction of the velocity. The negative sign indicates the force is resistive and opposes the motion. If the drag coefficient, C_d, is constant, then the drag force goes as the square of the speed. However, the drag coefficient is itself a complex function of velocity (*8*), and if there is a portion of the C_d *vs* v drag curve that behaves as $1/v$, then we have a net linear air-drag force (*9*). This is most easily expressed by saying that the drag coefficient goes as the inverse of the dimensionless Mach number, $\text{Ma} \equiv v/a$, where a is the speed of sound in air. In any case, it is useful to first solve the equations of motion for linear air resistance as a prelude to the more complex quadratic air drag.

So let us consider the case where $C_d \sim 1/\text{Ma}$ and let $C_d \implies C_d/\text{Ma}$. The vector equation of motion then has the form

$$\frac{d\mathbf{v}}{dt} = -b\mathbf{v} - g\mathbf{j}, \tag{14}$$

where the coefficient

$$b \equiv \frac{\rho a C_d A}{2m} \tag{15}$$

has dimensions of s^{-1}. In terms of the two components, the equations of motion are

$$\dot{u} = -bu \quad \text{and} \quad \dot{w} = -bw - g. \tag{16}$$

The solutions are

$$u = u_0 e^{-bt} \tag{17}$$

for the x component of the velocity and

$$x = \frac{u_0}{b}\left(1 - e^{-bt}\right) \tag{18}$$

for the downrange position (using the origin as the starting location).[1] The y component of the velocity is

$$w = \left(w_0 + \frac{g}{b}\right) e^{-bt} - \frac{g}{b} \tag{19}$$

and

$$y = \left(w_0 + \frac{g}{b}\right) \frac{1}{b}\left(1 - e^{-bt}\right) - \frac{gt}{b} \tag{20}$$

for the vertical position of the projectile (again using the origin as the starting position). Eliminating t between equations 18 and 20, gives the equation for the trajectory,

$$\boxed{y = \frac{g}{b^2}\left[\left(1 + \frac{bw_0}{g}\right)\frac{bx}{u_0} + \ln\left(1 - \frac{bx}{u_0}\right)\right]}. \tag{21}$$

It is instructive to compare this trajectory equation to its counterpart, which neglected air drag, equation 6. Absence of air drag results in y as a simple quadratic function of x. With linear air drag, x appears in a non-algebraic way, involving both x and its logarithm.

The range, R, is the distance downrange when $y = 0$, so it is the non-trivial solution to the equation

$$\left(1 + \frac{bw_0}{g}\right)\frac{bR}{u_0} + \ln\left(1 - \frac{bR}{u_0}\right) = 0. \tag{22}$$

This is a transcendental equation for R that, until recently, could not be solved algebraically but had to be solved numerically. With the definition of the Lambert W function[2] as the solution of

[1] Notice that combining equations 17 and 18 leads to the simple relationship $u = u_0 - bx$, which means that for shallow launch angles, the velocity is a linearly-decreasing function of range. This is a key observation of Celmins, who shows how this can be exploited to determine the drag coefficient (*9*).

[2] Corless, R. M.; Gonnet, G. H.; Hare, D. E. G.; Jeffrey, D. J.; Knuth, D. E. On the Lambert W function. *Advances in Computational Mathematics*, **1996**, *5* (1), 329-359. This paper is also available online at https://www.cs.uwaterloo.ca/research/tr/1993/03/W.pdf.

the equation

$$\boxed{W(z)e^{W(z)} = z}, \tag{23}$$

we now have a way of solving these types of equations (*6*, *10*). The strategy is to rearrange it into the form of equation 23 with a known quantity on the right-hand side to play the role of z, then we simply identify $W(z)$ as the Lambert W function.

We show the steps here. First, exponentiate both sides of equation 22 to give

$$\exp\left[\left(1+\frac{bw_0}{g}\right)\frac{bR}{u_0}\right]\left(1-\frac{bR}{u_0}\right) = 1, \tag{24}$$

and

$$\left(1+\frac{bw_0}{g}\right)\left(\frac{bR}{u_0}-1\right)\exp\left[\left(1+\frac{bw_0}{g}\right)\frac{bR}{u_0}\right] = -\left(1+\frac{bw_0}{g}\right), \tag{25}$$

and finally

$$\left(1+\frac{bw_0}{g}\right)\left(\frac{bR}{u_0}-1\right)\exp\left[\left(1+\frac{bw_0}{g}\right)\left(\frac{bR}{u_0}-1\right)\right] = -\left(1+\frac{bw_0}{g}\right)\exp\left[-\left(1+\frac{bw_0}{g}\right)\right]. \tag{26}$$

Comparing this to the Lambert equation, equation 23, we see that

$$W\left(-\left(1+\frac{bw_0}{g}\right)\exp\left[-\left(1+\frac{bw_0}{g}\right)\right]\right) = \left(1+\frac{bw_0}{g}\right)\left(\frac{bR}{u_0}-1\right). \tag{27}$$

Solving this for R gives

$$R = \frac{u_0}{b}\left[1+\frac{W(z)}{1+bw_0/g}\right], \tag{28}$$

or

$$\boxed{R = \frac{v_0\cos\theta}{b}\left[1+\frac{W(z)}{1+\alpha\sin\theta}\right]}, \tag{29}$$

where $W(z)$ is the Lambert function[3],

$$z \equiv -(1+\alpha\sin\theta)\exp[-(1+\alpha\sin\theta)] \quad \text{and} \quad \alpha \equiv bv_0/g. \tag{30}$$

3.1 Maximum Range

Now, we know that the angle that maximizes the range when there is no air resistance is 45°. To find the corresponding angle in the case of linear air resistance, we set the derivative of R with respect to θ equal to zero and solve for θ:

$$\left(\frac{dR}{d\theta}\right)_{\theta_{\max}} = 0.$$

[3]Since R must be positive, we require that $W(z) > -(1+\alpha\sin\theta)$ in equation 29, which in the limit as $\alpha \to 0$ requires $W(z) > -1$. This implies that here we want $W_0(z)$, the principal branch of the Lambert W function (see appendix A). Note that our focus has been on the range, but we haven't actually excluded the trivial solution of $x = 0$ when $y = 0$. Thus, if we use x rather than R in equation 29, this formula actually contains both solutions, $x = 0$ and $x = R$, due to the fact that $W(xe^x) = x$ is always the trivial solution for one of the branches. I am grateful to Dr. Joseph Collins for pointing this out.

Implicit differentiation of the Lambert equation, $We^W = z$, gives

$$\frac{dW}{dz} = \frac{e^{-W}}{1+W} = \frac{W}{(1+W)z}.$$

Using this, along with the chain rule and equation 30, we have

$$\frac{dW}{d\theta} = \frac{dW}{dz}\frac{dz}{d\theta} = \frac{W}{(1+W)z}\frac{dz}{d\theta} = \frac{W}{1+W}\frac{d\ln z}{d\theta} = \frac{W}{1+W}\frac{-\alpha^2\sin\theta\cos\theta}{1+\alpha\sin\theta}. \tag{31}$$

So, from equation 29, and using equation 31,

$$\begin{aligned}
\frac{dR}{d\theta} &= -\frac{v_0\sin\theta}{b}\left[1+\frac{W}{1+\alpha\sin\theta}\right] + \frac{v_0\cos\theta}{b}\frac{1}{1+\alpha\sin\theta}\frac{W}{1+W}\frac{-\alpha^2\sin\theta\cos\theta}{1+\alpha\sin\theta} + \frac{v_0\cos\theta}{b}W\frac{-\alpha\cos\theta}{(1+\alpha\sin\theta)^2} \\
&= -\frac{v_0\sin\theta}{b}\left[1+\frac{W}{1+\alpha\sin\theta}\right] - \frac{v_0\cos\theta}{b}\frac{W\alpha\cos\theta}{(1+\alpha\sin\theta)^2}\left[\frac{\alpha\sin\theta}{1+W}+1\right] \\
&= -\frac{v_0}{b}\frac{1+W+\alpha\sin\theta}{(1+\alpha\sin\theta)^2}\left[\sin\theta(1+\alpha\sin\theta) + \frac{W}{1+W}\alpha(1-\sin^2\theta)\right] \\
&= -\frac{v_0}{b}\frac{1+W+\alpha\sin\theta}{(1+\alpha\sin\theta)^2}\left[\frac{\alpha}{1+W}\sin^2\theta + \sin\theta + \frac{\alpha W}{1+W}\right] \\
&= -\frac{v_0}{b}\frac{1+W+\alpha\sin\theta}{(1+\alpha\sin\theta)^2}\frac{\alpha}{1+W}\left[\sin^2\theta + \frac{1+W}{\alpha}\sin\theta + W\right]
\end{aligned}$$

Requiring this to be zero gives the equation for θ_{max}, the firing angle to get the maximum range:

$$\sin^2\theta_{\text{max}} + \frac{1+W}{\alpha}\sin\theta_{\text{max}} + W = 0. \tag{32}$$

Notice that, since $W(z)$ is a function of θ (see equation 30), this is not a simple quadratic in $\sin\theta_{\text{max}}$. We can solve it numerically—for example, through bisection, since we know that the angle must lie somewhere in the interval $(0, \pi/4)$—but a better option is to, once again, make use of the Lambert function to solve it explicitly. First, let $x \equiv \sin\theta_{\text{max}}$, and solve equation 32 for W,

$$W(z) = -\sin\theta_{\text{max}}\frac{1+\alpha\sin\theta_{\text{max}}}{\alpha+\sin\theta_{\text{max}}} = -x\frac{1+\alpha x}{\alpha+x} \tag{33}$$

where, from equation 30, $z = -(1+\alpha x)e^{-(1+\alpha x)}$. From the defining relation of the Lambert function, equation 23, we know that

$$-x\frac{1+\alpha x}{\alpha+x}\exp\left[-x\frac{1+\alpha x}{\alpha+x}\right] = -(1+\alpha x)e^{-(1+\alpha x)}. \tag{34}$$

We want to solve this equation for x. Rearranging, we have

$$\frac{x}{x+\alpha}\exp\left[\frac{\alpha^2 x+\alpha}{x+\alpha}\right] = 1 \tag{35}$$

and

$$\frac{x}{x+\alpha}\exp\left[\frac{\alpha^2 x+x+\alpha-x}{x+\alpha}\right] = \frac{x}{x+\alpha}\exp\left[(\alpha^2-1)\frac{x}{x+\alpha}+1\right] = \frac{x}{x+\alpha}\exp\left[(\alpha^2-1)\frac{x}{x+\alpha}\right]e = 1,$$

so that

$$(\alpha^2-1)\frac{x}{x+\alpha}\exp\left[(\alpha^2-1)\frac{x}{x+\alpha}\right]=\frac{\alpha^2-1}{e},$$

and therefore,

$$W\left(\frac{\alpha^2-1}{e}\right)=(\alpha^2-1)\frac{x}{x+\alpha}.$$

Finally, solving this for $x=\sin\theta_{\text{max}}$, we get

$$\theta_{\text{max}}=\sin^{-1}\left[\frac{\alpha W\left(\frac{\alpha^2-1}{e}\right)}{\alpha^2-1-W\left(\frac{\alpha^2-1}{e}\right)}\right].$$

Since $W(0)=0$ (which follows from equation 23), as it stands this expression has the indeterminate form $0/0$ when $\alpha=1$. Returning to equation 35 and setting $\alpha=1$, we find that $x=1/(e-1)$. Therefore, the complete solution for the maximum range launch angle is

$$\boxed{\theta_{\text{max}}=\begin{cases}\sin^{-1}\left[\dfrac{\alpha W\left(\frac{\alpha^2-1}{e}\right)}{\alpha^2-1-W\left(\frac{\alpha^2-1}{e}\right)}\right] & \text{if } \alpha\neq 1\\[2ex] \sin^{-1}\left[\dfrac{1}{e-1}\right] & \text{if } \alpha=1\end{cases}} \tag{36}$$

and this is shown plotted in figure 2. The maximum range is obtained by substituting these values into equation 29 while making use of equation 33:

$$\boxed{R_{\text{max}}=\begin{cases}\dfrac{v_0^2}{g}\sqrt{\dfrac{\alpha^2-\left[1+W\left(\frac{\alpha^2-1}{e}\right)\right]^2}{\alpha^2(\alpha^2-1)}} & \text{if } \alpha\neq 1\\[2ex] \dfrac{v_0^2}{g}\sqrt{\dfrac{e-2}{e}} & \text{if } \alpha=1\end{cases}} \tag{37}$$

and where we used $v_0/b=(v_0^2/g)/\alpha$. Recall that the range at any angle θ in the case of no air resistance is $(v_0^2/g)\sin(2\theta)$. Let's now call this quantity R_0. Then, a natural comparison is R_{max}/R_0, which is the maximum range at the optimal launch angle *with air resistance*, as compared to the range at the same launch angle but *without air resistance*. Using equations 36 and 37, we get

$$\boxed{\frac{R_{\text{max}}}{R_0}=\begin{cases}\dfrac{\left[\alpha^2-1-W\left(\frac{\alpha^2-1}{e}\right)\right]^2}{2\alpha^2(\alpha^2-1)W\left(\frac{\alpha^2-1}{e}\right)} & \text{if } \alpha\neq 1\\[2ex] \dfrac{(e-1)^2}{2e} & \text{if } \alpha=1\end{cases}} \tag{38}$$

and this is shown plotted in figure 3. Both the ratio R_{max}/R_0 and θ_{max} are seen to be functions solely of the dimensionless parameter $\alpha\equiv bv_0/g$, which is itself the dimensionless ratio of the air

drag deceleration to the acceleration due to gravity.

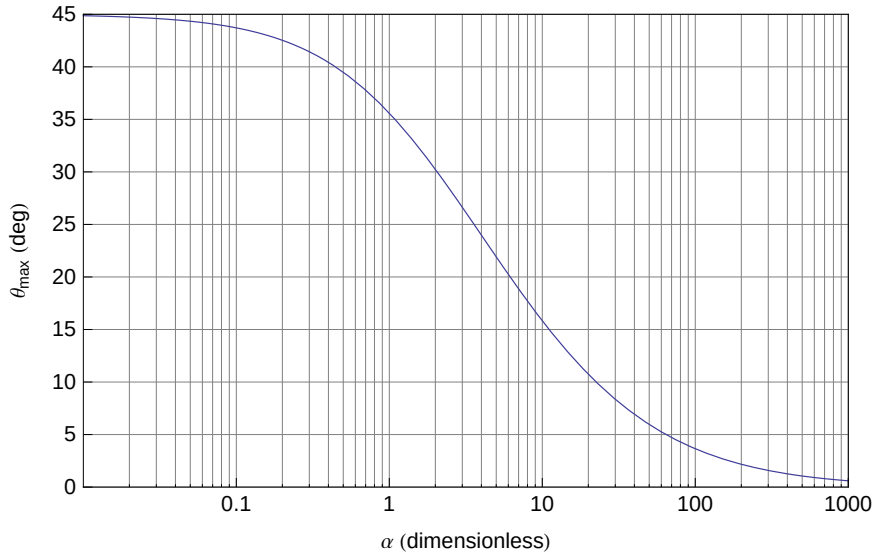

Figure 2. Firing angle to achieve maximum range as a function of the dimensionless parameter $\alpha = bv_0/g$.

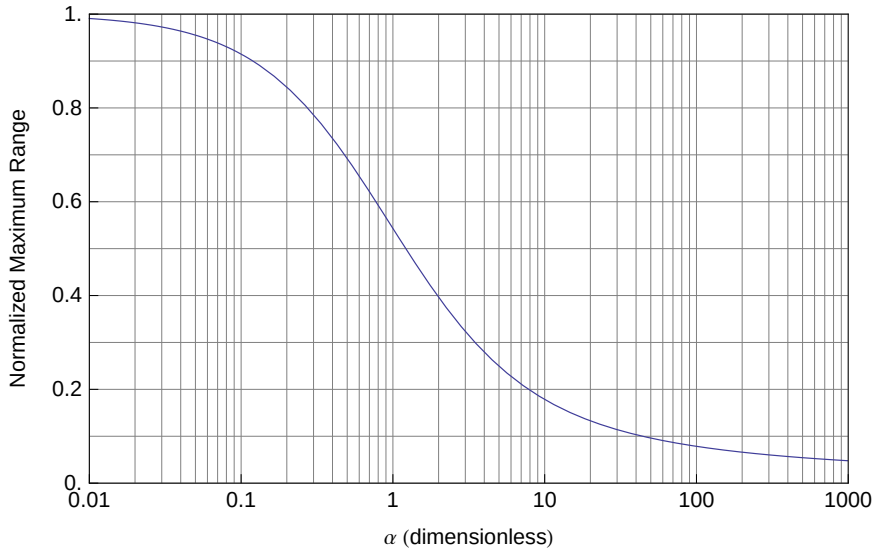

Figure 3. Normalized maximum range, $R_{\max}/R_0$, as a function of the dimensionless parameter $\alpha = bv_0/g$.

So, while 45° is the optimal launch angle when there is no air resistance, this is clearly not the case when air resistance is taken into account. Indeed, we see that the optimal angle is a rapidly decreasing function of α.

3.2 Trajectory Height

The maximum height of the trajectory occurs when w is momentarily zero, which, from equation 19, occurs when

$$t_{y=H} = \frac{1}{b}\ln(1+\alpha\sin\theta). \tag{39}$$

Substituting this time into equation 20, we get the trajectory height

$$\boxed{H = \frac{v_0\sin\theta}{b}\left[1 - \frac{\ln(1+\alpha\sin\theta)}{\alpha\sin\theta}\right]} \tag{40}$$

and, from equation 18, is located at the downrange distance

$$x_{y=H} = \frac{v_0\cos\theta}{b}\left(\frac{\alpha\sin\theta}{1+\alpha\sin\theta}\right). \tag{41}$$

3.3 Total Time of Flight and Average Speed

Using equations 18, 22, and 29, the total time of flight is

$$\boxed{T = \frac{1}{b}\left[1+\alpha\sin\theta+W(z)\right]}, \tag{42}$$

where

$$z = -(1+\alpha\sin\theta)\exp[-(1+\alpha\sin\theta)] \quad \text{and} \quad \alpha \equiv bv_0/g.$$

The average speed over the entire flight is $\bar{v} = R/T$, which gives

$$\boxed{\bar{v} = \frac{v_0\cos\theta}{1+\alpha\sin\theta}}. \tag{43}$$

3.4 Impact Velocity and Impact Angle

Impact occurs when $x = R$. The components of the impact velocity at this point are

$$\boxed{u = -u_0\frac{W(z)}{1+\alpha\sin\theta} \quad \text{and} \quad w = -w_0\frac{1+W(z)}{\alpha\sin\theta}}, \tag{44}$$

and the impact angle is

$$\boxed{\phi = \tan^{-1}\left[\left(\frac{1+\alpha\sin\theta}{\alpha\sin\theta}\right)\left(\frac{1+W(z)}{W(z)}\right)\tan\theta\right]}, \tag{45}$$

where θ is the launch angle, and again,

$$z = -(1+\alpha\sin\theta)\exp[-(1+\alpha\sin\theta)] \quad \text{and} \quad \alpha \equiv bv_0/g.$$

An interesting property of the maximum-range trajectory is that *the impact velocity is perpendicular to the launch velocity* (*11*). We can show this by demonstrating that the dot product of the two vectors vanishes. Using equation 44 and simplifying,

$$\begin{aligned}
\mathbf{v}_0 \cdot \mathbf{v} &= (u_0\mathbf{i} + w_0\mathbf{j}) \cdot \left(-u_0 \frac{W}{1+\alpha\sin\theta}\mathbf{i} - w_0 \frac{1+W}{\alpha\sin\theta}\mathbf{j} \right) \\
&= -v_0^2 \left[\frac{W(1-\sin^2\theta)}{1+\alpha\sin\theta} + \frac{(1+W)\sin^2\theta}{\alpha\sin\theta} \right] \\
&= -\frac{v_0^2}{1+\alpha\sin\theta} \left[W(1-\sin^2\theta) + \frac{(1+W)\sin\theta(1+\alpha\sin\theta)}{\alpha} \right] \\
&= -\frac{v_0^2}{1+\alpha\sin\theta} \left[\sin^2\theta + \frac{1+W}{\alpha}\sin\theta + W \right],
\end{aligned}$$

and from equation 32 the term in brackets equals zero when $\theta = \theta_{\text{max}}$. Thus, the impact angle in this case is given by $\phi_{\text{max}} = \theta_{\text{max}} - \pi/2$.[4]

The velocity components, u and w, are monotonically decreasing functions of both α and θ, although u decreases more rapidly than w. Consequently, as long as $\alpha > 0$, the impact angle is always greater (*i.e.*, steeper) than the launch angle. The functional relationship is shown in figures 4 and 5.

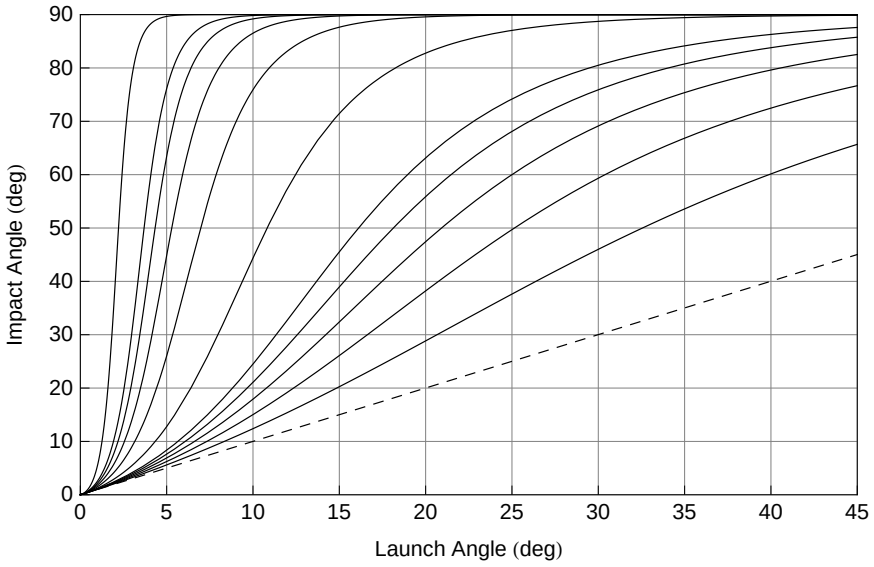

Figure 4. Impact angle, ϕ, as a function of launch angle, θ, for $\alpha = 1^\circ$, 2°, 3°, 4°, 5°, 10°, 20°, 30°, 40°, 50°, 100°. The dashed line is the limit as $\alpha \to 0$, which gives $\phi = \theta$.

[4]Notice that the impact velocity is also perpendicular to the launch velocity for the maximum range trajectory in the case of no air drag, since the launch angle is 45° and the impact angle is -45°.

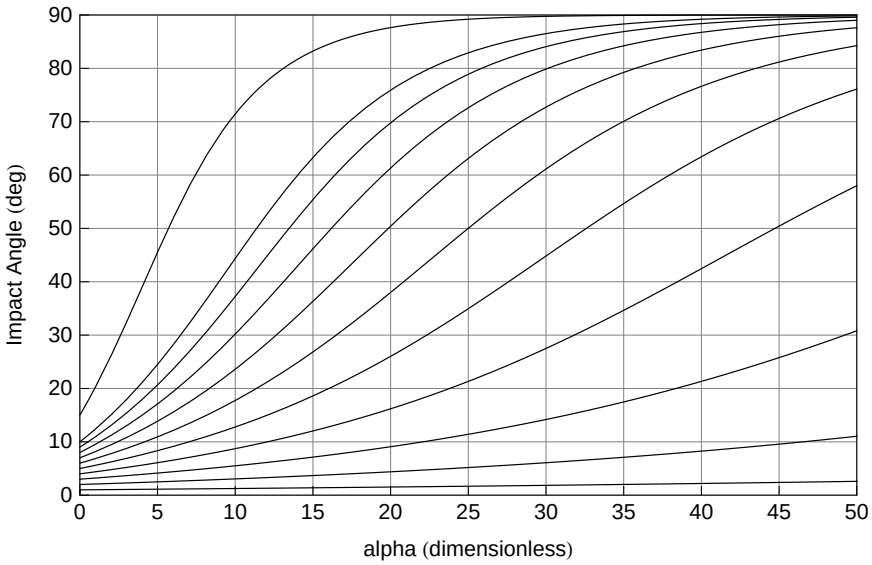

Figure 5. Impact angle as a function of α for launch angle $\theta = 0^\circ$ (lower curve), 1°, 2°, 3°, 4°, 5°, 6°, 7°, 8°, 9°, $10^\circ, 15^\circ$.

The contour plot in figure 6 shows the relationship between the launch angle, θ, and the dimensionless parameter α in order to achieve a given impact angle, ϕ. For example, if we want to achieve an impact angle of 80° or greater and $\alpha = 20$, then we need a launch angle $\theta \geq 11^\circ$, but if $\alpha = 30$, then $\theta \geq 8^\circ$.

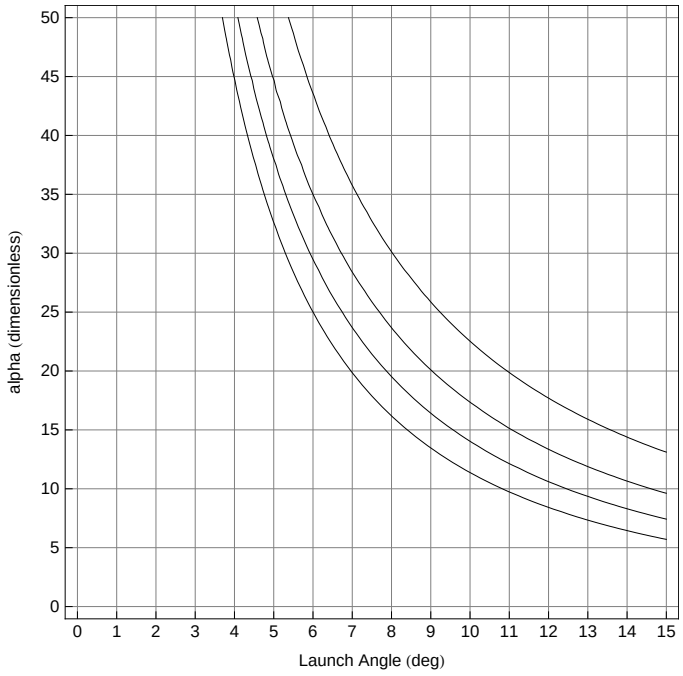

Figure 6. Contours of launch angle θ and α for which $\phi = 50^\circ$ (lower curve), 60°, 70°, 80°.

4. Projectile Motion with Quadratic Air Drag ($C_d \sim$ Constant)

We return to the quadratic drag force of equation 13, and now we assume a constant drag coefficient, C_d, independent of the Mach number. The total force acting on the projectile then is

$$\mathbf{F} = -\tfrac{1}{2} C_d A \rho v^2 \hat{\mathbf{v}} - mg\mathbf{j}, \tag{46}$$

and the equations of motion in terms of u and w are

$$\dot{u} = -bvu \quad \text{and} \quad \dot{w} = -bvw - g, \tag{47}$$

where

$$b \equiv \frac{\rho C_d A}{2m} \tag{48}$$

(which has dimensions of m^{-1} in SI units). Since $v = \sqrt{u^2 + w^2}$, the component equations are coupled, and the nature of this coupling prevents us from finding a general closed-form solution.[5] But if we limit ourselves to small launch angles, where $\tan\theta \ll 1$, then we can approximate $v = \sqrt{u^2 + w^2}$ with u, and this will permit a closed-form solution. This is the approximation that is analyzed throughout this section.

4.1 Flat-Fire Trajectory Approximation

Flat-fire trajectory (*8, 12*) is the name given to launch angles a few degress above the horizontal, usually less than 10°. In this case the horizontal velocity is on average much greater than the vertical velocity, $u \gg w$. Consequently, $v = \sqrt{u^2 + w^2} \approx u$, and the equations of motion become

$$\dot{u} = -bu^2 \quad \text{and} \quad \dot{w} = -buw - g. \tag{49}$$

The first equation is easily integrated to give the horizontal velocity

$$u = \frac{u_0}{1 + bu_0 t}. \tag{50}$$

Another integration gives x as a function of time,

$$x = \frac{1}{b} \ln(1 + bu_0 t), \tag{51}$$

where we are assuming the projectile is launched from the origin. To integrate the second equation of equation 49, let's first use the chain rule,

$$\frac{dw}{du}\frac{du}{dt} = \frac{dw}{dt},$$

and equation 49, to write

$$-bu^2 \frac{dw}{du} = -buw - g.$$

[5] It is not difficult to solve these equations numerically, and the standard technique using the fourth-order Runge-Kutta method is contained in the appendix B.

Then it is not difficult to rearrange this to

$$d\left(\frac{w}{u}\right) = -\frac{g}{2b}d\left(\frac{1}{u^2}\right),$$

from which the integration is now trivial, and solving for w gives

$$w = \frac{w_0 - gt/2}{1 + bu_0 t} - \frac{1}{2}gt. \tag{52}$$

Integrating again gives y as a function of time,

$$y = \left(w_0 + \frac{g}{2bu_0}\right)\frac{1}{bu_0}\ln(1 + bu_0 t) - \frac{gt}{2bu_0} - \frac{1}{4}gt^2. \tag{53}$$

Eliminating t between equations 51 and 53 gives the trajectory equation,

$$\boxed{y = \frac{g}{(2bu_0)^2}\left[(1 + \beta\sin 2\theta)2bx - (e^{2bx} - 1)\right]}, \tag{54}$$

where we have introduced the dimensionless parameter

$$\beta \equiv bv_0^2/g. \tag{55}$$

Once again, we see that accounting for air resistance results in a trajectory equation that involves x in a non-algebraic way. A plot of a typical trajectory is shown in figure 7. Due to the effect of air resistance, the path is no longer a parabola, nor is it symmetrical about the maximum height.

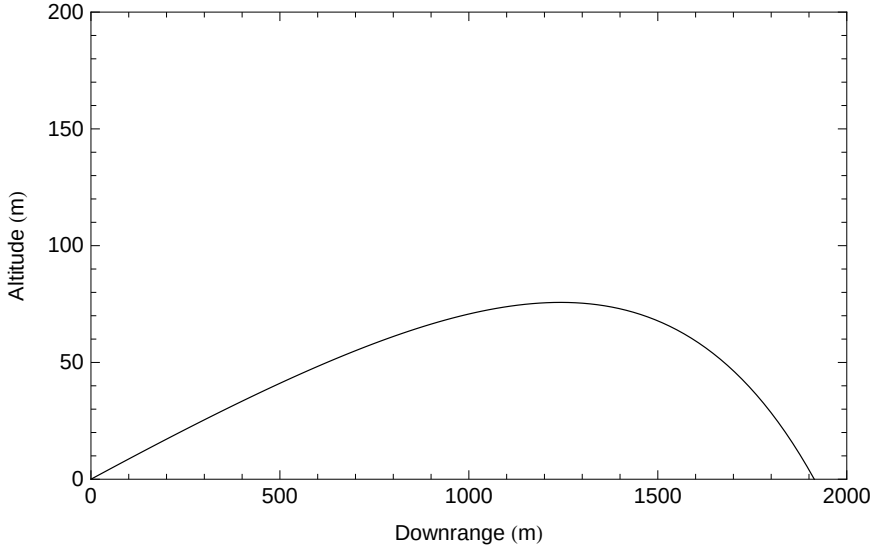

Figure 7. Flat-fire trajectory for a 5° firing angle and a 823 m/s muzzle velocity.

The range, R, is the non-trivial value of x that gives $y = 0$, so that

$$(1 + \beta\sin 2\theta)2bR - (e^{2bR} - 1) = 0. \tag{56}$$

By a now-familiar routine, we solve this equation for R by rearranging it into the Lambert W functional form. The steps are as follows. Multiply through by e^{-2bR} and rearrange to get

$$[(1+\beta\sin 2\theta)2bR+1]e^{-2bR}=1, \tag{57}$$

or

$$\left[-2bR-\frac{1}{1+\beta\sin 2\theta}\right]e^{-2bR}=-\frac{1}{1+\beta\sin 2\theta}.$$

Finally, multiply through by $\exp[-(1+\beta\sin 2\theta)^{-1}]$ to get

$$\left[-2bR-\frac{1}{1+\beta\sin 2\theta}\right]\exp\left(-2bR-\frac{1}{1+\beta\sin 2\theta}\right)=-\frac{1}{1+\beta\sin 2\theta}\exp\left(-\frac{1}{1+\beta\sin 2\theta}\right).$$

We recognize this equation to be of the form $W(z)e^{W(z)}=z$ with

$$z\equiv-(1+\beta\sin 2\theta)^{-1}\exp\left[-(1+\beta\sin 2\theta)^{-1}\right],$$

a known quantity, and

$$W(z)=-2bR-\frac{1}{1+\beta\sin 2\theta}.$$

Therefore, the range is given by

$$\boxed{R=-\frac{1}{2b}\left[W(z)+\frac{1}{1+\beta\sin 2\theta}\right]}, \tag{58}$$

where $W(z)$ is the Lambert function[6],

$$z\equiv-(1+\beta\sin 2\theta)^{-1}\exp\left[-(1+\beta\sin 2\theta)^{-1}\right]\quad\text{and}\quad\beta\equiv bv_0^2/g. \tag{59}$$

Notice that the firing angle θ enters into the range formula only through the particular combination $\beta\sin 2\theta$.[7] Furthermore, notice that the firing angle dependence only involves the combination of $\sin 2\theta$, the same as the case of no air resistance. This would imply that the optimum angle is again 45°. But this has no practical significance since we must restrict the launch angle to be small for the approximation to be valid in the first place!

Figure 8 shows how the range depends upon the firing velocity for a fixed firing angle.

[6]Since R must be positive, we require that $W(z)<-(1+\beta\sin 2\theta)^{-1}$ in equation 58, which in the limit as $\beta\to 0$ requires $W(z)<-1$. This implies that here we want $W_{-1}(z)$, the secondary branch of the Lambert W function, which varies from -1 to $-\infty$ as z varies from $-1/e$ to 0 (see appendix A). In Mathematica®, the principal branch is known as `LambertW[z]` and the secondary branch is known as `LambertW[-1,z]`. C++ code for both branches is listed in appendix B.

[7]Contrast this with the case of linear air resistance, equation 29, where the firing angle enters into the range formula outside of α.

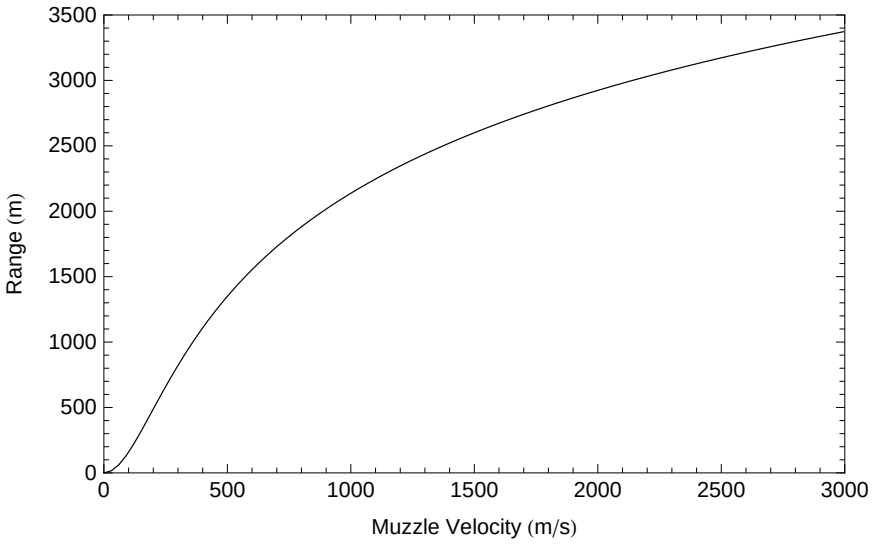

Figure 8. Range as a function of muzzle velocity for a 5° firing angle.

4.2 Trajectory Height

The maximum height of the trajectory, H, can be obtained by setting $dy/dx = 0$, solving for x, and substituting back into the formula for y. This gives

$$\boxed{H = \frac{\tan\theta}{2b}\left[\frac{1+\beta\sin 2\theta}{\beta\sin 2\theta}\ln(1+\beta\sin 2\theta) - 1\right]} \tag{60}$$

and occurs when

$$t_{y=H} = \frac{\sqrt{1+\beta\sin\theta} - 1}{bv_0\cos\theta}.$$

The maximum height is located at the downrange distance

$$x_{y=H} = \frac{1}{2b}\ln(1+\beta\sin 2\theta).$$

4.3 Total Time of Flight and Average Speed

The total time of flight, T, is

$$\boxed{T = \frac{\sqrt{-W(z)(1+\beta\sin 2\theta)} - 1}{bv_0\cos\theta}}, \tag{61}$$

where we used equations 51, 57, and 58.

The average speed over the entire flight is $\bar{v} = R/T$, and we find

$$\boxed{\bar{v} = v_0\cos\theta\frac{\sqrt{-W(z)(1+\beta\sin 2\theta)} + 1}{2(1+\beta\sin 2\theta)}}. \tag{62}$$

4.4 Impact Velocity and Impact Angle

Let us also calculate the velocity of the projectile as a function of its distance downrange. From the chain rule,

$$\frac{du}{dx} = \frac{\dot{u}}{u} \quad \text{and} \quad \frac{dw}{dx} = \frac{\dot{w}}{u}. \tag{63}$$

Using equation 49, we get

$$\frac{du}{dx} = -bu \quad \text{and} \quad \frac{dw}{dx} = -bw - \frac{g}{u}.$$

The first of these is easily integrated,

$$u = u_0 e^{-bx}, \tag{64}$$

and when substituted into the second, we get the differential equation

$$\frac{dw}{dx} + bw = -\frac{g}{u_0} e^{bx}.$$

Multiplying through by e^{bx} gives

$$e^{bx}\frac{dw}{dx} + be^{bx}w = \frac{d}{dx}(we^{bx}) = -\frac{g}{u_0}e^{2bx},$$

which is easily integrated to give

$$w = w_0 e^{-bx} - \frac{g}{2bu_0}(e^{bx} - e^{-bx}). \tag{65}$$

At impact, $x = R$, and we get

$$\boxed{u = \frac{v_0 \cos\theta}{\sqrt{-W(z)(1+\beta\sin 2\theta)}}} \tag{66}$$

for the horizontal component and

$$\boxed{w = \frac{v_0 \sin\theta}{\sqrt{-W(z)(1+\beta\sin 2\theta)}}\left[1 + \frac{W(z)(1+\beta\sin 2\theta)+1}{\beta \sin 2\theta}\right]} \tag{67}$$

for the vertical component of the impact velocity. The impact angle, ϕ, is given by

$$\boxed{\phi = \tan^{-1}\left[\left(1 + \frac{W(z)(1+\beta\sin 2\theta)+1}{\beta\sin 2\theta}\right)\tan\theta\right]}, \tag{68}$$

where

$$z = -(1+\beta\sin 2\theta)^{-1}\exp\left[-(1+\beta\sin 2\theta)^{-1}\right] \quad \text{and} \quad \beta = bv_0^2/g.$$

The impact angle is plotted in figure 9 as a function of the launch angle for various values of β. As in the case of linear air resistance, the impact angle is always greater (steeper) than the launch angle as long as $\beta > 0$. However, we see that in the case of quadratic resistance, the effect is not quite so pronounced as that of linear air resistance (*cf.* figure 4).

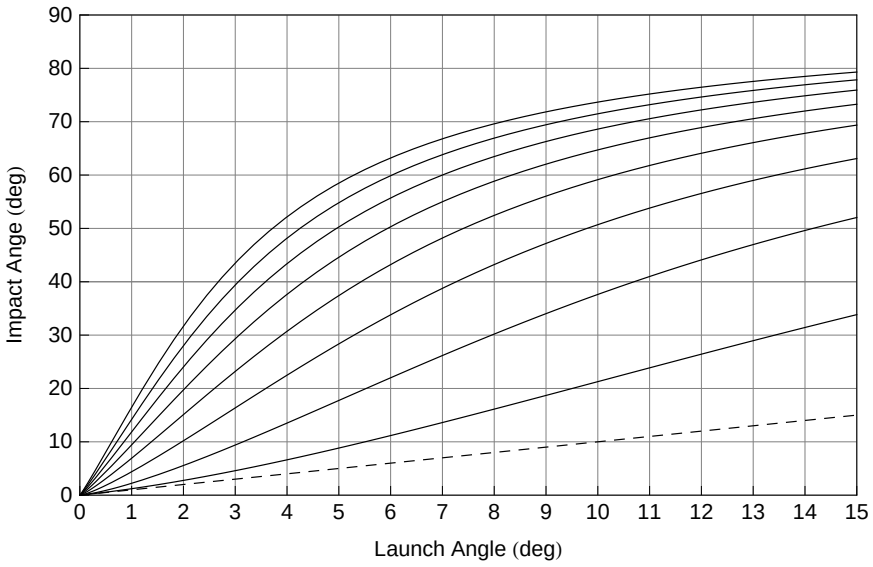

Figure 9. Impact angle as a function of launch angle θ for $\beta = 0$ (dashed), 10, 10^2, 10^3, 10^4, 10^5, 10^6, 10^7, 10^8.

5. Discussion

In general, the drag coefficient is a complex function of both the Reynold's number, Re, and the Mach number, Ma (*8*, *13*). In the case of exterior ballistics through the air that we are considering here, Re $> 10^5$ and in this regime, the drag coefficient tends to be insensitive to the Reynold's number and becomes a function of Mach number only (*14*, *15*).

The drag coefficient for cubes and spheres have been studied extensively and drag curves have been fitted (*16*, *17*) to the following equations:

$$C_d(\mathrm{Ma})_{\mathrm{sphere}} = \begin{cases} 0.45\mathrm{Ma}^2 + 0.424 & \text{if } 0 \leq \mathrm{Ma} \leq 0.722 \\ 2.1e^{-1.2(\mathrm{Ma}+0.35)} - 8.9e^{-2.2(\mathrm{Ma}+0.35)} + 0.92 & \text{if } \mathrm{Ma} > 0.722 \end{cases} \tag{69}$$

$$C_d(\mathrm{Ma})_{\mathrm{cube}} = \begin{cases} 0.60\mathrm{Ma}^2 + 1.04 & \text{if } 0 \leq \mathrm{Ma} \leq 1.131 \\ 2.1e^{-1.16(\mathrm{Ma}+0.35)} - 6.5e^{-2.23(\mathrm{Ma}+0.35)} + 1.67 & \text{if } \mathrm{Ma} > 1.131 \end{cases} \tag{70}$$

The drag curves are shown in figure 10. Increasing the speed through the speed of sound (Mach I) results in a steep increase in the drag coefficient, it reaches a peak value slightly beyond Mach I, and then it gradually decreases in value as the speed continues to increase. Drag curves for bullets depend upon the nose and shape and show greater variation than the curves in figure 10 (*8*).

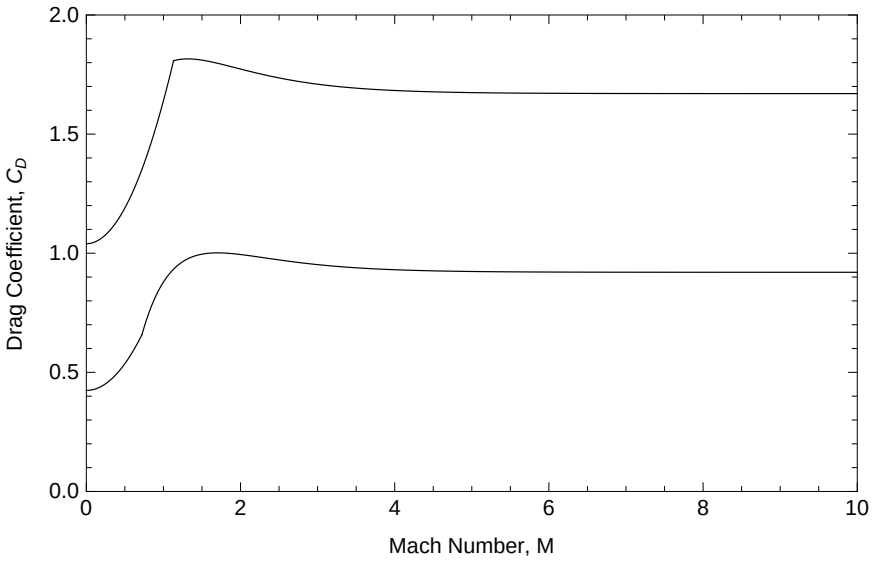

Figure 10. Drag coefficient as a function of Mach number for cube (upper curve) and sphere (lower).

5.1 Measuring the Drag Coefficient

We saw that in the case of linear air resistance, the horizontal component of the velocity falls off linearly with distance downrange: $u = u_0 - bx$. Now, if we restrict ourselves to flat-fire trajectories, then we can neglect w as compared to u, and $v \approx u$. Then it follows from equations 17, 18, 50, and 51 that

$$v = v_0 - bx \quad \text{and} \quad \ln v = \ln v_0 - bt \tag{71}$$

if and only if $C_d \sim 1/\text{Ma}$, and

$$\ln v = \ln v_0 - bx \quad \text{and} \quad \frac{1}{v} = \frac{1}{v_0} + bt \tag{72}$$

if and only if $C_d \sim$ constant. Thus, if we have velocity break screens positioned at various locations downrange and record the velocity at each one, along with the downrange distance x and the time of impact t, then these equations tell us how to find the dimensionless coefficient b from the slope of the straight line. Once we have determined b, we can calculate C_d from equations 15 and 48.

A more systematic approach for treating a general drag curve is contained in the reports by Weinacht, Cooper, and Newill (*18, 19*).

5.2 Sample Case

Let's consider a sample case of a 1-kg steel sphere with a constant drag coefficient of $C_d = 0.5$ and a launch speed of 1000 m/s. In the case of linear air resistance, we find that the angle for maximum range is $\theta_{\text{max}} = 8°$ and $R_{\text{max}} = 3111$ m. This is shown in figure 11, where it is clear that $45°$ no longer gives the maximum range. Also notice how quickly the impact angle approaches $90°$ as the launch angle increases. The value of b for this example can be calculated from equation

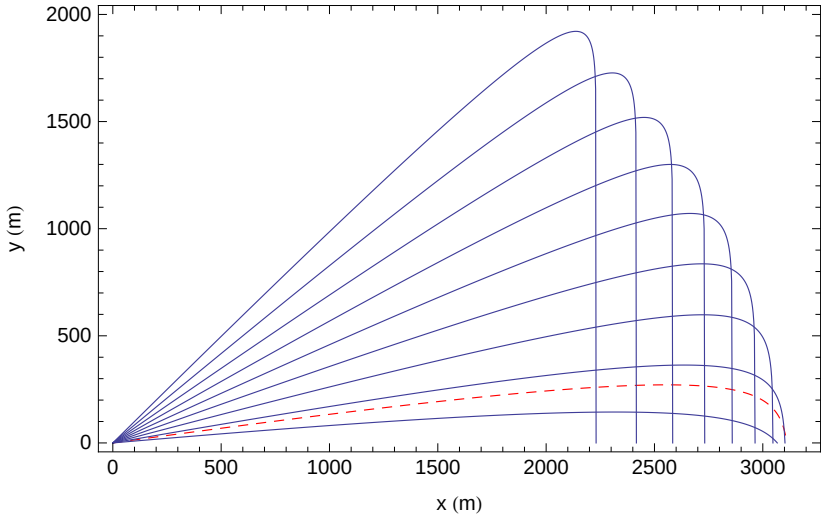

Figure 11. Linear air drag trajectories for $\theta = 5°, 10°, 15°, 20°, 25°, 30°, 35°, 40°, 45°$ (solid blue) and $\theta_{\max} = 8°$ (dashed red).

15 and we find $b = 0.317\text{s}^{-1}$, which means that the air drag deceleration is initially $bv_0 = 317$ m/s^2, or about 32 g's.

In the case of quadratic air resistance, we could only solve the equations exactly when we made the approximation of shallow launch angles (*flat-fire approximation*). The resulting equations lead to $\theta = 45°$ as the optimal angle, but this would invalidate the approximation that $\tan\theta \ll 1$. Under these circumstances, we are not in a position to optimize the range. However, we can ask how well our flat-fire approximate trajectories compare to those from the Runge-Kutta solution (*20*) of the exact quadratic air drag problem, and this is shown in figure 12.

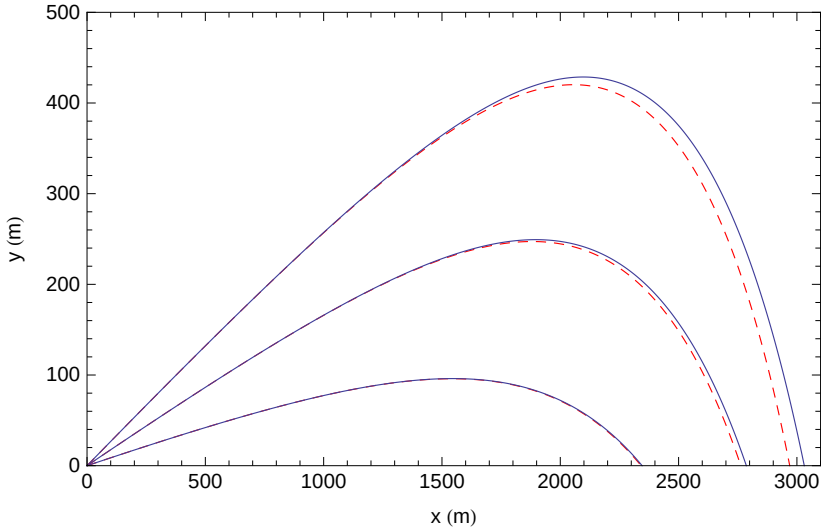

Figure 12. Quadratic air drag trajectories in the flat-fire approximation (solid blue) compared to Runge-Kutta solutions (dashed red) for $\theta = 5°, 10°, 15°$.

The approximate closed-form solution overpredicts the range by only 0.2% at 5° and by less than 1% at 10°. Even at 15°, it overpredicts the range by only 2%. The value of b for this example can be calculated from equation 46 and we get $b = 9.24 \times 10^4$ m^{-1}, which means that the air drag deceleration is initially $bv_0^2 = 924$ m/s^2, or about 94 g's.

To find the launch angle that will result in the maximum range, we made a number of Runge-Kutta runs. Figure 13 shows that in this case the maximum range is obtained when the launch angle is 25°.

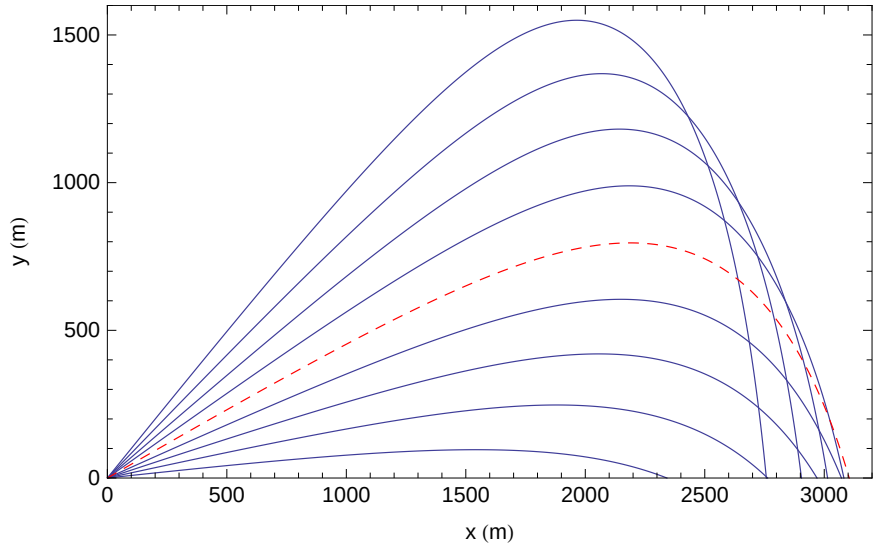

Figure 13. Quadratic air drag trajectories using Runge-Kutta for $\theta = 5^\circ$, 10°, 15°, 20°, 30°, 35°, 40°, and 45° (solid blue) with maximum range at 25° (dashed red).

6. Conclusion

The Lambert W function permits us to solve certain types of transcendental equations algebraically. With its aid, we are able to derive exact analytical expressions for projectile motion in the case of linear air drag and approximate analytical expressions in the case of quadratic air drag. Much like the case of no air resistance, the linear air drag case permits a complete analytical solution, including the optimum launch angle that maximizes the range, along with the maximum range itself. Furthermore, we demonstrated that a general property of the maximum range trajectory is that the impact velocity is perpendicular to the launch velocity. This applies to all values of the dimensionless parameter α; in particular, it applies to the limit as $\alpha \to 0$, which is the case of no air resistance.

The approximate solution in the case of quadratic air drag only applies to shallow launch angles, but we showed that this is a good approximation for angles less than about 10° to 15° by comparing it to the Runge-Kutta numerical solution.

In general the drag coefficient is a complex function of the Mach number. However, depending upon the launch and impact speeds, it may well be the case that the drag coefficient takes on the simple form $C_d \sim 1/\text{Ma}$ or $C_d \sim$ constant, in which case the linear or quadratic air drag equations apply. By making use of the average speed equations over the trajectory (equations 43 and 62) we are able to check whether either condition applies. Finally, equations 71 and 72 provide an empirical method for computing the drag coefficient in the two regimes considered in this report.

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Appendix A. Plot of the Lambert W Function

First consider the equation $y = xe^x$, which is shown plotted in figure A–1. It is defined for $-\infty < x < +\infty$ and has an absolute minimum of $y = -1/e$ at $x = -1$. As $x \to -\infty$, $y \to 0^-$, and as $x \to +\infty$, $y \to +\infty$. Substituting a value for x, we can readily calculate a value for y.

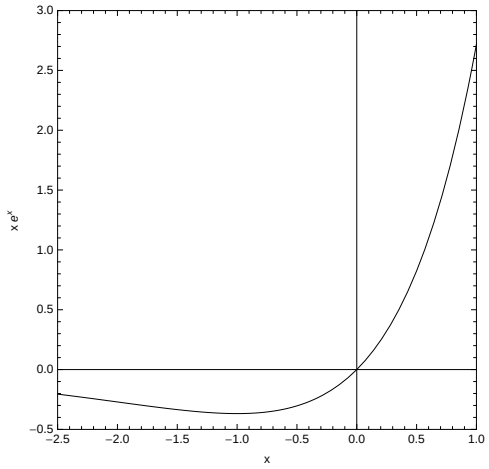

Figure A–1. Plot of $y = xe^x$.

Now consider the inverse, $ye^y = x$. For a given value of x, the solution to this equation is $y = W(x)$, where $W(x)$ is the Lambert function, and is shown plotted in figure A–2.

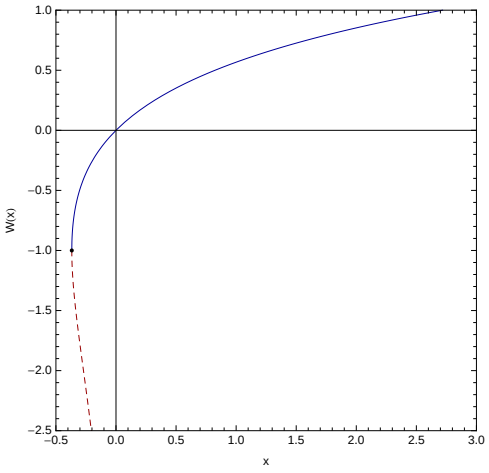

Figure A–2. Plot of $W(x)$.

There are two real branches to the function: $W_0(x)$, which is shown in solid blue, is defined on the half-open interval $[-1/e, +\infty)$, where $W_0(x)$ varies from -1 to $+\infty$; and $W_{-1}(x)$, which is shown in dashed red, is defined on the half-open interval $[-1/e, 0)$, where $W_{-1}(x)$ varies from -1 to $-\infty$.

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Appendix B. C++ Source Code Listings

The C++ source code listing for both branches of the Lambert W function is as follows (*5*):

Listing 1. lambertW.h

```
// lambertW.h: header file for both branches of the Lambert W function
// Ref: "Having Fun with Lambert W(x) Function," Darko Veberic, arXiv:1003.1628v1 [cs.MS], 8 Mar 2010
// R. Saucier, July 2012

#ifndef LAMBERT_H
#define LAMBERT_H

#include <iostream>
#include <cstdlib>
#include <cmath>
using namespace std;

// principal branch; domain is half-open interval [-1/e, infinity) and range is half-open interval [-1, infinity)
double lambertW_0( double z ) {

   const double P[10] = { -1., 1., -1./3., 11./72., -43./540., 769./17280., -221./8505., -680863./43545600.,
                          -1963./204120., 226287557./37623398400. };
   const double A[5]  = { 1., 5.931375839364438, 11.39220550532913, 7.33888339911111, 0.653449016991959 };
   const double B[5]  = { 1., 6.931373689597704, 16.82349461388016, 16.43072324143226, 5.115235195211697 };
   const double C[5]  = { 1., 2.445053070726557, 1.343664225958226, 0.148440055397592, 0.0008047501729130 };
   const double D[5]  = { 1., 3.444708986486002, 3.292489857371952, 0.916460018803122, 0.0530686404483322 };

   double p, a, a2, a3, a4, a5, b, b2, b3, b4;
   if ( -1. / M_E <= z && z < -0.32358170806015724 ) {
      p = sqrt( 2. * ( 1. + M_E * z ) );
      return P[0] +
             p * ( P[1] +
             p * ( P[2] +
             p * ( P[3] +
             p * ( P[4] +
             p * ( P[5] +
             p * ( P[6] +
             p * ( P[7] +
             p * ( P[8] +
             p * ( P[9] ) ) ) ) ) ) ) ) );
   }
   else if ( -0.32358170806015724 <= z && z < 0.14546954290661823 ) {
      return z * ( A[0] +
             z * ( A[1] +
             z * ( A[2] +
             z * ( A[3] +
             z * ( A[4] ) ) ) ) ) / ( B[0] + z * ( B[1] +
                                      z * ( B[2] +
                                      z * ( B[3] +
                                      z * ( B[4] ) ) ) ) );
   }
   else if ( 0.14546954290661823 <= z && z < 8.706658967856612 ) {
      return z * ( C[0] +
             z * ( C[1] +
             z * ( C[2] +
             z * ( C[3] +
             z * ( C[4] ) ) ) ) ) / ( D[0] + z * ( D[1] +
                                      z * ( D[2] +
                                      z * ( D[3] +
                                      z * ( D[4] ) ) ) ) );
   }
   else if ( 8.706658967856612 <= z ) {
      a = log( z );
      a2 = a * a;
      a3 = a * a2;
      a4 = a * a3;
      a5 = a * a4;
      b = log( log( z ) );
      b2 = b * b;
      b3 = b * b2;
```

```
      b4 = b * b3;
      return a - b + b / a + ( -2. + b ) * b / ( 2. * a2 ) + ( 6. - 9. * b + 2. * b2 ) * b / ( 6. * a3 ) +
             ( -12. + 36. * b - 22. * b2 + 3. * b3 ) * b / ( 12. * a4 ) +
             ( 60. - 300. * b + 350. * b2 - 125. * b3 + 12. * b4 ) * b / ( 60. * a5 );
   }
   else {
      cerr << "Error: z = " << z << " outside domain of lambertW_0(z)" << endl << "Program stopped" << endl;
      exit( EXIT_FAILURE );
   }
}

// secondary branch; domain is half-open interval [-1/e, 0) and range is half-open interval (-infinity, -1]
double lambertW_1( double z ) {

   const double P[10] = { -1., 1., -1./3., 11./72., -43./540., 769./17280., -221./8505., -680863./43545600., -1963./204120.,
                          226287557./37623398400. };
   const double A[3]  = { -7.81417672390744, 253.88810188892484, 657.9493176902304 };
   const double B[6]  = { 1., -60.43958713690808, 99.9856708310761, 682.6073999909428, 962.1784396969866, 1477.9341280760887 };

   double p, a, a2, a3, a4, a5, b, b2, b3, b4;
   if ( -1. / M_E <= z && z < -0.30298541769 ) {
      p = -sqrt( 2. * ( 1. + M_E * z ) );
      return P[0] +
             p * ( P[1] +
             p * ( P[2] +
             p * ( P[3] +
             p * ( P[4] +
             p * ( P[5] +
             p * ( P[6] +
             p * ( P[7] +
             p * ( P[8] +
             p * ( P[9] ) ) ) ) ) ) ) ) );
   }
   else if ( -0.30298541769 <= z && z < -0.051012917658221676 ) {
      return ( A[0] +
               z * ( A[1] +
               z * ( A[2] ) ) ) / ( B[0] + z * ( B[1] +
                                          z * ( B[2] +
                                          z * ( B[3] +
                                          z * ( B[4] +
                                          z * ( B[5] ) ) ) ) ) );
   }
   else if ( -0.051012917658221676 < z && z < 0 ) {
      a = log( -z );
      a2 = a * a;
      a3 = a * a2;
      a4 = a * a3;
      a5 = a * a4;
      b = log( -log( -z ) );
      b2 = b * b;
      b3 = b * b2;
      b4 = b * b3;
      return a - b + b / a + ( -2. + b ) * b / ( 2. * a2 ) + ( 6. - 9. * b + 2. * b2 ) * b / ( 6. * a3 ) +
             ( -12. + 36. * b - 22. * b2 + 3. * b3 ) * b / ( 12. * a4 ) +
             ( 60. - 300. * b + 350. * b2 - 125. * b3 + 12. * b4 ) * b / ( 60. * a5 );
   }
   else {
      cerr << "Error: z = " << z << " outside domain of lambertW_1(z)" << endl << "Program stopped" << endl;
      exit( EXIT_FAILURE );
   }
}
#endif
```

The following is a simple main program that makes use of the function:

Listing 2. lambertW.C

```
// lambertW.C: Fast evaluation of both branches of the Lambert W function over its full domain [-1/e,infinity)
// Ref: "Having Fun with Lambert W(x) Function," Darko Veberic, arXiv:1003.1628v1 [cs.MS], 8 Mar 2010
// R. Saucier, July 2012

#include "lambertW.h"
#include <iostream>
#include <iomanip>
#include <cstdlib>
#include <cmath>
using namespace std;

int main ( int argc, char* argv[] ) {

   const double Z_MIN = -1. / M_E;

   double z = Z_MIN;
   if ( argc == 2 ) z = atof( argv[1] );

   if ( Z_MIN <= z && z < 0. ) {   // both branches are define in this domain
      cout << "z = " << z << " lambertW_0( z ) = " << lambertW_0( z ) << endl;
      cout << "z = " << z << " lambertW_1( z ) = " << lambertW_1( z ) << endl;
   }
   else if ( 0. <= z ) {   // only principal branch is defined in this domain
      cout << "z = " << z << " lambertW_0( z ) = " << lambertW_0( z ) << endl;
   }
   else {   // outside domain of both branches
     cerr << "Outside domain of function; program stopped" << endl;
     exit( EXIT_FAILURE );
   }

        return EXIT_SUCCESS;
}
```

Here are some sample evaluations:

Listing 3. Sample Evaluations

```
./lambertW
z = -0.367879 lambertW_0( z ) = -1
z = -0.367879 lambertW_1( z ) = -1

./lambertW -0.15
z = -0.15 lambertW_0( z ) = -0.179491
z = -0.15 lambertW_1( z ) = -2.99359

./lambertW 1.7
z = 1.7 lambertW_0( z ) = 0.779601
```

The following is the implementation of the linear air resistance formulas contained in section 3:

Listing 4. linear.C

```
// linear.C: Projectile motion with air resistance linearly dependent on velocity (SI Units)
// R. Saucier, July 2012

#include "lambertW.h"
#include <fstream>
#include <iostream>
#include <cmath>
#include <cstdlib>
using namespace std;

double y( double G, double B, double v0, double th, double x  ) {   // trajectory: y as a function of x

   double u0 = v0 * cos( th );
   double w0 = v0 * sin( th );
   double z  = B * x / u0;
   return ( G / ( B * B ) ) * ( ( 1. + B * w0 / G ) * z + log( 1. - z ) );
}

double th_max( double G, double B, double v0  ) {   // launch angle for max range (rad)

   double alpha = B * v0 / G;
   if ( alpha == 1. ) return asin( 1. / ( M_E - 1. ) );
   double a = ( alpha - 1. ) * ( alpha + 1. );
   double W = lambertW_0( a / M_E );
   return asin( alpha * W / ( a - W ) );
}

double range_max( double G, double B, double v0  ) {   // max range over all launch angles (m)

   double c = v0 * v0 / G;
   double alpha = B * v0 / G;
   if ( alpha == 1. ) return c * sqrt( ( M_E - 2. ) / M_E );
   double a = ( alpha - 1. ) * ( alpha + 1. );
   double W = lambertW_0( a / M_E );
   return c * sqrt( ( alpha - ( 1. + W ) ) * ( alpha + ( 1. + W ) ) / ( alpha * alpha * a ) );
}

double range_norm( double alpha ){   // normalized max range (dimensionless)

   if ( alpha == 1. ) return ( M_E - 1. ) * ( M_E - 1. ) / ( 2. * M_E );
   double a = ( alpha - 1. ) * ( alpha + 1. );
   double W = lambertW_0( a / M_E );
   return ( a - W ) * ( a - W ) / ( 2. * alpha * alpha * a * W );
}

double range( double G, double B, double v0, double th ) {   // range at the given launch angle (m)

   double u0    = v0 * cos( th );
   double alpha = B * v0 / G;
   double a     = 1. + alpha * sin( th );
   double z     = -a * exp( -a );
   return ( u0 / B ) * ( 1. + lambertW_0( z ) / a );
}

double height( double G, double B, double v0, double th ) {   // max y over the trajectory (m)

    double w0    = v0 * sin( th );
    double alpha = B * v0 / G;
    double a     = alpha * sin( th );
    return ( w0 / B ) * ( 1. - log( 1. + a ) / a );
}

double t_height( double G, double B, double v0, double th  ) {   // time to reach max height of trajectory (m)

   double alpha = B * v0 / G;
   return log( 1. + alpha * sin( th ) ) / B;
}

double x_height( double G, double B, double v0, double th  ) {   // x distance when at max height (m)

   double u0    = v0 * cos( th );
```

```
   double alpha = B * v0 / G;
   double a     = alpha * sin( th );
   return ( u0 / B ) * a / ( 1. + a );
}

double t_range( double G, double B, double v0, double th  ) {   // total time of flight (s)

   double alpha = B * v0 / G;
   double a     = 1. + alpha * sin( th );
   double z = -a * exp( -a );
   return  ( a + lambertW_0( z ) ) / B;
}

double v_bar( double G, double B, double v0, double th  ) {   // average speed over entire flight (s)

   double u0    = v0 * cos( th );
   double alpha = B * v0 / G;
   double a     = 1. + alpha * sin( th );
   return  u0 / a;
}

double u_impact( double G, double B, double v0, double th ) {   // x-component of velocity at impact (m/s)

   double u0    = v0 * cos( th );
   double alpha = B * v0 / G;
   double a     = 1. + alpha * sin( th );
   double z     = -a * exp( -a );
   double W     = lambertW_0( z );
   return -u0 * W / a;
}

double w_impact( double G, double B, double v0, double th ) {   // y-component of velocity at impact (m/s)

   double w0    = v0 * sin( th );
   double alpha = B * v0 / G;
   double a     = 1. + alpha * sin( th );
   double z     = -a * exp( -a );
   double W     = lambertW_0( z );
   return -w0 * ( 1. + W ) / ( a - 1. );
}

double v_impact( double G, double B, double v0, double th ) {   // speed at impact (m/s)

  double u = u_impact( G, B, v0, th );
  double w = w_impact( G, B, v0, th );
  return sqrt( u * u + w * w );
}

double th_impact( double G, double b, double v0, double th ) {   // angle at impact (rad)

   double alpha = b * v0 / G;
   double a     = 1. + alpha * sin( th );
   double z     = -a * exp( -a );
   double W     = lambertW_0( z );
   return atan( ( a / ( a - 1. ) ) * ( ( 1. + W ) / W ) * tan( th ) );
}

double u( double B, double v0, double th, double t ) {   // x-component of velocity at time t (m/s)

   return v0 * cos( th ) * exp( -B * t );
}

double w( double G, double B, double v0, double th, double t ) {   // y-component of velocity at time t (m/s)

   return ( v0 * sin( th ) + G / B ) * exp( -B * t ) - G / B;
}

int main( int argc, char* argv[] ) {

   const double D2R     = M_PI / 180.;                    // to convert degrees to radians
   const double R2D     = 180. / M_PI;                    // to convert radians to degrees
   const double G       = 9.80665;                        // acceleration due to gravity (m/s^2)
   const double RHO_AIR = 1.205;                          // density of air at 20 deg Celsius (kg/m^3)
   const double V_SOUND = 343.2;                          // speed of sound in air at 20 deg Celsius (m/s)
   const double m       = 1.;                             // mass of projectile (kg)
   const double rho     = 7830.;                          // density of steel (kg/m^3)
```

```
   const double D       = pow( 6. * m / ( M_PI * rho ), 1./3. );      // diameter of spherical projectile (m)
   const double Ap      = 0.25 * M_PI * D * D;                        // presented area of spherical projectile (m^2)
   const double Cd      = 0.5;                                        // drag coefficient (dimensionless)
   const double B       = RHO_AIR * V_SOUND * Cd * Ap / ( 2. * m );   // lumped drag coefficient (s^(-1))

   double v0 = 1000.;                        // launch speed (m/s)
   double deg = th_max( G, B, v0 ) * R2D;    // default launch angle is for max range (deg)
   if ( argc == 2 ) deg = atof( argv[1] );   // launch angle specified on command line (deg)
   double th = deg * D2R;                    // launch angle converted to rad (rad)

   double alpha = B * v0 / G;                // dimensionless ratio of deceleration due to air drag and acceleration due to gravity
   double R     = range( G, B, v0, th );     // range at given launch angle (m)

   cout << "Launch angle (deg)                    = " << deg                          << endl
        << "b (1/s)                               = " << B                            << endl
        << "alpha (-)                             = " << alpha                        << endl
        << "Optimal angle for max range (deg)     = " << th_max( G, B, v0 ) * R2D     << endl
        << "Max range (m)                         = " << range_max( G, B, v0 )        << endl
        << "Normalized max range, Rmax/R0 (-)     = " << range_norm( alpha )          << endl
        << "Range at given launch angle (m)       = " << R                            << endl
        << "Maximum height of trajectory (m)      = " << height( G, B, v0, th )       << endl
        << "Elapsed time to achieve max height (s) = " << t_height( G, B, v0, th )    << endl
        << "Downrange distance at max height (m)  = " << x_height( G, B, v0, th )     << endl
        << "Total time of flight (s)              = " << t_range( G, B, v0, th )      << endl
        << "Average speed over entire flight (m/s) = " << v_bar( G, B, v0, th )       << endl
        << "Impact speed (m/s)                    = " << v_impact( G, B, v0, th )     << endl
        << "Impact angle (deg)                    = " << th_impact( G, B, v0, th ) * R2D << endl;

   ofstream fout( "linear.out" );
   for ( double x = 0.; x <= R; x += R / 999. ) fout << x << "\t" << y( G, B, v0, th, x ) << endl;
   fout << R << "\t" << 0. << endl;
   fout.close();

   return EXIT_SUCCESS;
}
```

The following is the implementation of the quadratic air drag formulas contained in section 4:

Listing 5. quadratic.C

```
// quadratic.C: Projectile motion with air resistance quadratically dependent on velocity (SI Units)
// R. Saucier, July 2012

#include "lambertW.h"
#include <fstream>
#include <iostream>
#include <cmath>
#include <cstdlib>
#include <cassert>
using namespace std;

inline double sqr( double x ) { return x * x; }

double y( double G, double B, double v0, double th, double x ) {   // trajectory: y as a function of x (m)

   double u0   = v0 * cos( th );
   double beta = B * v0 * v0 / G;
   double a    = 1. + beta * sin( 2. * th );
   double z    = 2. * B * x;
   return ( G / sqr( 2. * B * u0 ) ) * ( a * z - ( exp( z ) - 1. ) );
}

double range( double G, double B, double v0, double th ) {   // range (m)

   double beta = B * v0 * v0 / G;
   double zeta = 1. / ( 1. + beta * sin( 2. * th ) );
   double z    = -zeta * exp( -zeta );
   return -( lambertW_1( z ) + zeta ) / ( 2. * B );
}

double height( double G, double B, double v0, double th ) {   // height of trajectory (m)

   double beta = B * v0 * v0 / G;
```

```
   double z    = 1. + beta * sin( 2. * th );
   return ( tan( th ) / ( 2. * B ) ) * ( z * log( z ) / ( z - 1. ) - 1. );
}

double t_height( double G, double B, double v0, double th ) {   // time to reach max height of trajectory (m)

   double beta = B * v0 * v0 / G;
   return ( sqrt( 1. + beta * sin( 2. * th ) ) - 1. ) / ( B * v0 * cos( th ) );
}

double x_height( double G, double B, double v0, double th ) {   // downrange distance at max height of trajectory (m)

   double beta = B * v0 * v0 / G;
   double z    = 1. + beta * sin( 2. * th );
   return log( z ) / ( 2. * B );
}

double t_range( double G, double B, double v0, double th  ) {   // total time of flight

   double beta = B * v0 * v0 / G;
   double zeta = 1. + beta * sin( 2. * th );
   double a    = 1. / zeta;
   double z    = -a * exp( -a );
   double W    = lambertW_1( z );
   return ( sqrt( -W * zeta ) - 1. ) / ( B * v0 * cos( th ) );
}

double v_bar( double G, double B, double v0, double th  ) {   // average speed over entire flight

   double u0   = v0 * cos( th );
   double beta = B * v0 * v0 / G;
   double zeta = 1. + beta * sin( 2. * th );
   double a    = 1. / zeta;
   double z    = -a * exp( -a );
   double W    = lambertW_1( z );
   return u0 * ( sqrt( -W * zeta ) + 1. ) / ( 2. * zeta );
}

double u_impact( double G, double B, double v0, double th ) {   // x-component of velocity at impact (m/s)

   double u0   = v0 * cos( th );
   double beta = B * v0 * v0 / G;
   double zeta = 1. + beta * sin( 2. * th );
   double a    = 1. / zeta;
   double z    = -a * exp( -a );
   double W    = lambertW_1( z );
   return u0 / sqrt( -W * zeta );
}

double w_impact( double G, double B, double v0, double th ) {   // y-component of velocity at impact (m/s)

   double w0   = v0 * sin( th );
   double beta = B * v0 * v0 / G;
   double zeta = 1. + beta * sin( 2. * th );
   double a    = 1. / zeta;
   double z    = -a * exp( -a );
   double W    = lambertW_1( z );
   return w0 * ( 1. + ( W * zeta + 1. ) / ( zeta - 1. ) ) / sqrt( -W * zeta );
}

double v_impact( double G, double B, double v0, double th ) {   // speed at impact (m/s)

   double u = u_impact( G, B, v0, th );
   double w = w_impact( G, B, v0, th );
   return sqrt( u * u + w * w );
}

double th_impact( double G, double B, double v0, double th ) {   // angle at impact (rad)

   double beta = B * v0 * v0 / G;
   double zeta = 1. + beta * sin( 2. * th );
   double a    = 1. / zeta;
   double z    = -a * exp( -a );
   double W    = lambertW_1( z );
   return atan( ( 1. + ( W * zeta + 1. ) / ( zeta - 1. ) ) * tan( th ) );
}
```

```
double u( double B, double v0, double th, double t ) {   // x-component of velocity (m/s)

   double u0 = v0 * cos( th );
   return u0 / ( 1. + B * u0 * t );
}

double x_time( double B, double v0, double th, double t ) {   // distance downrange (m)

   double u0 = v0 * cos( th );
   return log( 1. + B * u0 * t ) / B;
}

double w( double G, double B, double v0, double th, double t ) {   // y-component of velocity (m/s)

   double u0 = v0 * cos( th );
   double w0 = v0 * sin( th );
   double c  = 0.5 * G * t;
   return ( w0 - c ) / ( 1. + B * u0 * t ) - c;
}

double y_time( double G, double B, double v0, double th, double t ) {   // height (m)

   double u0 = v0 * cos( th );
   double w0 = v0 * sin( th );
   double c  = G  / ( 2. * B * u0 );
   return ( w0 + c ) * log( 1. + B * u0 * t ) / ( B * u0 ) - c * t - 0.25 * G * t * t;
}

int main( int argc, char* argv[] ) {

   const double D2R     = M_PI / 180.;                          // to convert degrees to radians
   const double R2D     = 180. / M_PI;                          // to convert radians to degrees
   const double G       = 9.80665;                              // acceleration due to gravity (m/s^2)
   const double RHO_AIR = 1.205;                                // density of air at 20 deg Celsius (kg/m^3)

   const double m       = 1.;                                   // mass of projectile (kg)
   const double rho     = 7830.;                                // density of steel (kg/m^3)
   const double D       = pow( 6. * m / ( M_PI * rho ), 1./3. );// diameter of spherical projectile (m)
   const double Ap      = 0.25 * M_PI * D * D;                  // presented area of spherical projectile (m^2)
   const double Cd      = 0.5;                                  // drag coefficient (dimensionless)
   const double B       = RHO_AIR * Cd * Ap / ( 2. * m );       // m^-1

   double v0 = 1000.;                       // launch speed (m/s)
   double deg = 5.;                         // default launch angle (deg)
   if ( argc == 2 ) deg = atof( argv[1] );  // launch angle specified on command line (deg)
   double th = deg * D2R;                   // launch angle converted to rad (rad)

   double beta = B * v0 * v0 / G;           // dimensionless ratio of deceleration due to air drag and acceleration due to gravity
   double R = range( G, B, v0, th );        // range (m)
   double T = t_range( G, B, v0, th  );     // total time of flight (s)

   cout << "Launch angle (deg)                     = " << deg                          << endl
        << "b (m^-1)                               = " << B                            << endl
        << "beta (-)                               = " << beta                         << endl
        << "Range of trajectory (m)                = " << R                            << endl
        << "Maximum height of trajectory (m)       = " << height( G, B, v0, th )       << endl
        << "Elapsed time to achieve max height (s) = " << t_height( G, B, v0, th )     << endl
        << "Downrange distance at max height (m)   = " << x_height( G, B, v0, th )     << endl
        << "Total time of flight (s)               = " << T                            << endl
        << "Average speed over entire flight (m/s) = " << v_bar( G, B, v0, th )        << endl
        << "Impact speed (m/s)                     = " << v_impact( G, B, v0, th )     << endl
        << "Impact angle (deg)                     = " << th_impact( G, B, v0, th ) * R2D << endl;

   ofstream fout( "quadratic.out" );
   for ( double x = 0.; x < R; x += R / 999. ) fout << x << "   " << y( G, B, v0, th, x ) << endl;
   fout << R << "\t" << 0. << endl;
   fout.close();

   return EXIT_SUCCESS;
}
```

The following is the implementation of the Runge-Kutta (*3, 20*) program used to test the approximate closed-form solution discussed in section 5:

Listing 6. rk.C

```
// rk.C: Runge–Kutta fourth–order method for projectile motion under influence of gravity and air resistance
//       Adapted from Richard Sandmeyer's program (SI Units)
// R. Saucier, July 2012

#include <iostream>
#include <fstream>
#include <cstdlib>
#include <cmath>
using namespace std;

int main( int argc, char* argv[] ) {

   const double D2R     = M_PI / 180.;   // to convert deg to rad
   const double R2D     = 1. / D2R;      // to convert rad to deg
   const double G       = 9.80665;       // acceleration due to gravity at sea level (m/s^2)
   const double RHO_AIR = 1.205;         // air density at 20 deg Celsius (kg/m^3)
   const double h       = 0.0005;        // time step (s)
   const double XOUT    = 1.;            // output approximately every 1 m downrange
   const double YSTOP   = 0.;            // cutoff altitude (m)
   double rho, m, b, beta, c, v, xnew, u, unew, xout, y, ynew, wnew, w, d, cd, ap, x, t, deg, theta;
   double k1x, k2x, k3x, k4x, l1x, l2x, l3x, l4x, k1y, k2y, k3y, k4y, l1y, l2y, l3y, l4y;

   rho   = 7830.;                                  // density of steel (kg/m^3)
   m     = 1.;                                     // mass of projectile (kg)
   d     = pow( 6. * m / ( M_PI * rho ), 1./3. );  // diameter of spherical projectile (m)
   ap    = 0.25 * M_PI * d * d;                    // presented area of spherical projectile (m^2)
   cd    = 0.5;                                    // drag coefficient (dimensionless)
   v     = 1000.;                                  // initial velocity (m/s)
   deg   = 5.;                                     // default launch angle (deg)
   if ( argc == 2 ) deg = atof( argv[1] );         // launch angle specified on command line (deg)
   theta = deg * D2R;                              // convert launch angle from deg to radians

   u = v * cos( theta );   // x component of velocity (m/s)
   w = v * sin( theta );   // y component of velocity (m/s)
   x     = 0.;
   y     = 0.;
   t     = 0.;
   xout  = XOUT;

   // fourth–order Runge–Kutta method inside this "while" loop (fixed step size, not adaptive step size)

   double height = 0., t_height = 0., x_height = 0.;
   b    = RHO_AIR * cd * ap / ( 2. * m );
   beta = b * v * v / G;

   ofstream fout( "rk.out" );
   while ( y > -0.001 ) {

      k1x  = h * u;
      k1y  = h * w;
      v    = sqrt( u * u + w * w );
      c    = -b * v;
      l1x  = h * ( c * u );
      l1y  = h * ( c * w - G );

      k2x  = h * ( u + 0.5 * l1x );
      k2y  = h * ( w + 0.5 * l1y );
      unew = u + 0.5 * l1x;
      wnew = w + 0.5 * l1y;
      v    = sqrt( unew * unew + wnew * wnew );
      c    = -b * v;
      l2x  = h * ( c * unew );
      l2y  = h * ( c * wnew - G );

      k3x  = h * ( u + 0.5 * l2x );
      k3y  = h * ( w + 0.5 * l2y );
      unew = u + 0.5 * l2x;
      wnew = w + 0.5 * l2y;
      v    = sqrt( unew * unew + wnew * wnew );
```

```
      c     = -b * v;
      l3x   = h * ( c * unew );
      l3y   = h * ( c * wnew - G );

      k4x   = h * ( u + l3x );
      k4y   = h * ( w + l3y );
      unew = u + l3x;
      wnew = w + l3y;
      v     = sqrt( unew * unew + wnew * wnew );
      c     = -b * v;
      l4x   = h * ( c * unew );
      l4y   = h * ( c * wnew - G );

      xnew = x + ( k1x + k2x + k2x + k3x + k3x + k4x ) / 6.;
      ynew = y + ( k1y + k2y + k2y + k3y + k3y + k4y ) / 6.;
      unew = u + ( l1x + l2x + l2x + l3x + l3x + l4x ) / 6.;
      wnew = w + ( l1y + l2y + l2y + l3y + l3y + l4y ) / 6.;

      t += h;
      x  = xnew;
      y  = ynew;
      u  = unew;
      w  = wnew;
      v  = sqrt( u * u + w * w );

      if ( y > height ) {
         height   = y;
         t_height = t;
         x_height = x;
      }

      if ( y <= YSTOP ) {
         x -= y * u / w;
         y  = YSTOP;
         fout << x << "\t" << y << endl;
         break;
      }

      if ( x >= xout ) {
         fout << x << "\t" << y << endl;
         xout += XOUT;
      }
   }
   fout.close();

   cout << "Launch angle (deg)                    = " << deg                  << endl
        << "b (m^-1)                              = " << b                    << endl
        << "beta (-)                              = " << beta                 << endl
        << "Range of trajectory (m)               = " << x                    << endl
        << "Maximum height of trajectory (m)      = " << height               << endl
        << "Elapsed time to achieve max height (s) = " << t_height            << endl
        << "Downrange distance at max height (m)  = " << x_height             << endl
        << "Total time of flight (s)              = " << t                    << endl
        << "Average speed over entire flight (m/s) = " << x / t               << endl
        << "Impact speed (m/s)                    = " << sqrt( u * u + w * w ) << endl
        << "Impact angle (deg)                    = " << atan( w / u ) * R2D  << endl;

   return EXIT_SUCCESS;
}
```

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